TO: ALL HOLDERS OF OFF WING ESCAPE SYSTEM INTEGRATOR ASSEMBLY COMPONENT MAINTENANCE MANUAL 25-65-62

REVISION NO. 8 DATED JUL 01/98

HIGHLIGHTS

Pages which have been added or revised are outlined below together with the highlights of the revision. Remove and insert the affected pages as listed and enter Revision No. and date on the Record of Revision Sheet.

CHAPTER/SECTION AND PAGE NO.	DESCRIPTION OF CHANGE
301	Changed part number callout of ball (75, IPL Fig. 2),
1002,1004	and deleted ball (77).

BOEING

COMPONENT
MAINTENANCE MANUAL

OFF WING ESCAPE SYSTEM INTEGRATOR ASSEMBLY

PART NUMBERS 416T2811-5 THRU -14
015T0449-4 THRU -7

COMPONENT MAINTENANCE MANUAL
WITH
ILLUSTRATED PARTS LIST

144222

BOEING
COMPONENT
MAINTENANCE MANUAL

REVISION RECORD

- Retain this record in front of manual. On receipt of revision, insert revised pages in the manual, and enter revision number, date inserted and initial.

REVISION NUMBER	REVISION DATE	DATE FILED	BY	REVISION NUMBER	REVISION DATE	DATE FILED	BY

144223

BOEING

COMPONENT
MAINTENANCE MANUAL

TEMPORARY REVISION AND SERVICE BULLETIN RECORD

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVE	DATE OF INCORPORATION INTO MANUAL
25-51		PRR B11170	OCT 10/85
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25A0104		PRR B11786	JUL 01/91
25A0104, R1			JUN 01/95
25A0104, R2			JUN 01/95

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25-65-62		
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1	JUN 01/95	01.1
2	BLANK	
REVISION RECORD		
1	JUL 01/91	01.1
2	BLANK	
TR & SB RECORD		
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2	BLANK	
LIST OF EFFECTIVE PAGES		
*1	JUL 01/98	01
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*301	JUL 01/98	01.1
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701	JUL 01/91	01.1
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ILLUSTRATED PARTS LIST		
1001	JUL 01/91	01.1
*1002	JUL 01/98	01.1
1003	JUN 01/95	01.1
*1004	JUL 01/98	01.1
1005	JUN 01/95	01.1
1006	JUN 01/95	01.1

PAGE	DATE	CODE

* = REVISED, ADDED OR DELETED

TABLE OF CONTENTS

*[1] Special instructions not required. Use standard industry practices.

INTRODUCTION

The instructions in this manual provide the information necessary to perform maintenance functions ranging from simple checks and replacement to complete shop-type repair.

This manual is divided into separate sections:

1. Title Page
2. Record of Revisions
3. Temporary Revision & Service Bulletin Record
4. List of Effective Pages
5. Table of Contents
6. Introduction
7. Procedures & IPL Sections

Refer to the Table of Contents for the page location of applicable sections. An asterisked flagnote *[ ] in place of the page number indicates that no special instructions are provided since the function can be performed using standard industry practices.

The beginning of the REPAIR section includes a list of the separate repairs, a list of applicable standard Boeing practices.

An explanation of the use of the Illustrated Parts List is provided in the Introduction to that section.

All weights and measurements used in the manual are in English units, unless otherwise stated. When metric equivalents are given they will be in parentheses following the English units.

Design changes, optional parts, configuration differences and Service Bulletin modifications create alternate part numbers. These are identified in the Illustrated Parts List (IPL) by adding an alphabetical character to the basic item number. The resulting item number is called an alpha-variant. Throughout the manual, IPL basic item number references also apply to alpha-variants unless otherwise indicated.

Verification:
Disassembly -- Oct 30/87
Assembly -- Oct 30/87

OFF-WING ESCAPE SYSTEM INTEGRATOR ASSEMBLY
DESCRIPTION AND OPERATION

1. The off-wing escape system integrator assembly consists of body, connector, hook, carrier, bellcrank, and fasteners. The integrator converts the emergency firing of a pyrotechnic actuator into a deployment of the off-wing escape slide.

2. Leading Particulars (approximate)

   Length -- 7 inches
   Width -- 3 inches
   Height -- 3 inches
   Weight -- 1.5 pounds

DISASSEMBLY

1. Remove screws (10) and plate (15) from body (120).

2. Lift out connector (20) and hook (30) from body (120).

3. Drive out pin (35) holding cam (55) at bellcrank (40) end opposite crank attach end. Slip freed cam (55) off bellcrank; then slide bellcrank out of body (120). Drive out remaining pin (35) and separate other cam (55) from bellcrank (40).

4. Remove screw (65), spring (70) and ball (75) from body (120).

5. Remove screws (80), lock plate (85) and latch lock (90) from body (120).

6. Do not remove crosspin (50) unless necessary for replacement of carrier (100), body (120), or crosspin (50).

CHECK

NOTE: Refer to IPL Fig. 1 for item numbers.

1. Check all parts in accordance with standard industry practices.

2. Magnetic particle check per 20-20-01 -- Plate (15), pawl (26), pins (27, 28), connector (29), hook (30), bellcrank (40, 45), cam (55, 60), lockplate (85), carrier (100, 105, 110, 115), body (120).

REPAIR - GENERAL

1. Content

A. Repair, refinish and replacement procedures are included in separate repair sections as follows:

P/N	NAME	REPAIR
416T2812-9,-34	CROSS PIN	1-1
416T2812-10,-33	LOCK SUPPORT PLATE	2-1
416T2813	BODY	3-1
416T2814	CARRIER	4-1
416T2815	CONNECTOR	5-1
416T2817	CAM	6-1
416T2818	BELLCRANK	7-1
416T2819	LOCK	8-1
416T2823	CAP PLATE	9-1
- - -	MISCELLANEOUS PARTS REFINISH	10-1

2. Standard Practices

A. Refer to the following standard practices, as applicable, for details of procedures in individual repairs:

20-30-02 Stripping of Protective Finishes
20-30-03 General Cleaning Procedures
20-41-01 Decoding Table for Boeing Finish Codes
20-42-03 Hard Chrome Plating
20-50-08 Application of Bonded Solid Film Lubricant
20-60-02 Finishing Materials

3. Materials

NOTE: Equivalent substitutes can be used.

A. Primer -- BMS 10-11, Type 1 color yellow (Ref 20-60-02)

B. Enamel -- BMS 10-11, Type 2 color white (Ref 20-60-02)

C. Fluorescent paint -- Zynolyte 1419, color red-orange (Ref 20-60-02)

D. Protective finish -- Type 41 (Ref 20-60-02)

E. Solid film lube -- BMS 3-8 (Ref 20-50-08)

4. Dimensioning Symbols

A. Standard True Position Dimensioning Symbols used in applicable repair procedures are shown in Fig. 601.

—	STRAIGHTNESS
▱	FLATNESS
⊥	PERPENDICULARITY (OR SQUARENESS)
//	PARALLELISM
○	ROUNDNESS
⌭	CYLINDRICITY
⌒	PROFILE OF A LINE
⌓	PROFILE OF A SURFACE
◎	CONCENTRICITY
⌯	SYMMETRY
∠	ANGULARITY
↗	RUNOUT
⌰	TOTAL RUNOUT
⌴	COUNTERBORE OR SPOTFACE
∨	COUNTERSINK
⌖	THEORETICAL EXACT POSITION OF A FEATURE (TRUE POSITION)
∅	DIAMETER
S ∅	SPHERICAL DIAMETER
R	RADIUS
SR	SPHERICAL RADIUS
()	REFERENCE
BASIC (BSC) OR [DIM]	A THEORETICALLY EXACT DIMENSION USED TO DESCRIBE SIZE, SHAPE OR LOCATION OF A FEATURE FROM WHICH PERMISSIBLE VARIATIONS ARE ESTABLISHED BY TOLERANCES ON OTHER DIMENSIONS OR NOTES.
[-A-]	DATUM
Ⓜ	MAXIMUM MATERIAL CONDITION (MMC)
Ⓛ	LEAST MATERIAL CONDITION (LMC)
Ⓢ	REGARDLESS OF FEATURE SIZE (RFS)
Ⓟ	PROJECTED TOLERANCE ZONE
FIM	FULL INDICATOR MOVEMENT

EXAMPLES

[— | 0.002]	STRAIGHT WITHIN 0.002
[⊥ | 0.002 | B]	PERPENDICULAR TO B WITHIN 0.002
[// | 0.002 | A]	PARALLEL TO A WITHIN 0.002
[○ | 0.002]	ROUND WITHIN 0.002
[⌭ | 0.010]	CYLINDRICAL SURFACE MUST LIE BETWEEN TWO CONCENTRIC CYLINDERS, ONE OF WHICH HAS A RADIUS 0.010 INCH GREATER THAN THE OTHER
[⌒ | 0.006 | A]	EACH LINE ELEMENT OF THE SURFACE AT ANY CROSS SECTION MUST LIE BETWEEN TWO PROFILE BOUNDARIES 0.006 INCH APART RELATIVE TO DATUM PLANE A
[⌓ | 0.020 | A]	SURFACES MUST LIE WITHIN PARALLEL BOUNDARIES 0.02 INCH APART AND EQUALLY DISPOSED ABOUT TRUE PROFILE
[◎ | ∅ 0.0005 | C]	CONCENTRIC TO C WITHIN 0.0005 DIAMETER
[⌯ | 0.010 | A]	SYMMETRICAL WITH A WITHIN 0.010
[∠ | 0.005 | A]	ANGULAR TOLERANCE 0.005 WITH A
[⌖ | ∅ 0.002 Ⓢ | B]	LOCATED AT TRUE POSITION WITHIN 0.002 DIA RELATIVE TO DATUM B, REGARDLESS OF FEATURE SIZE
[⊥ | ∅ 0.010 Ⓜ | A] [0.510 Ⓟ]	AXIS IS TOTALLY WITHIN A CYLINDER OF 0.010-INCH DIAMETER, PERPENDICULAR TO, AND EXTENDING 0.510-INCH ABOVE, DATUM A, MAXIMUM MATERIAL CONDITION
[2.000] OR 2.000 BSC	THEORETICALLY EXACT DIMENSION IS 2.000

NOTE: DATUM MAY APPEAR AT EITHER SIDE OF TOLERANCE FRAME [0.020 | A] [A | 0.020]

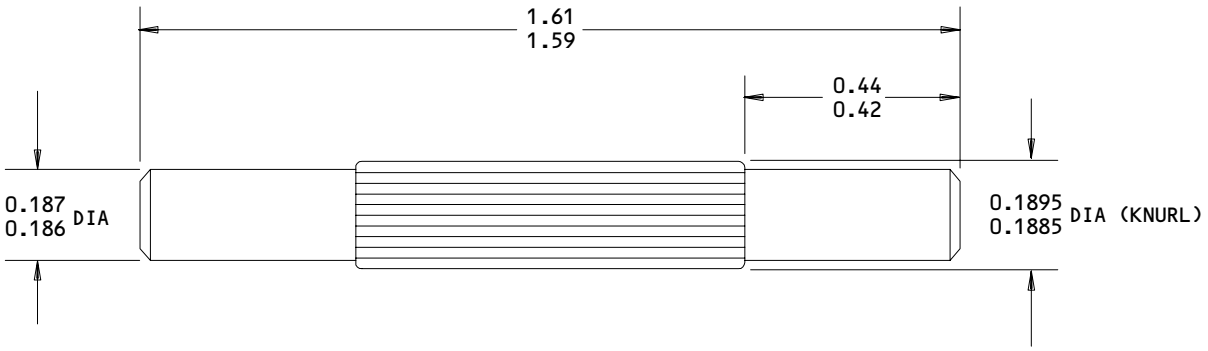
True Position Dimensioning Symbols
Figure 601

CROSS PIN - REPAIR 1-1

416T2812-9, -34

1. Plating Repair

A. Repair is only replacement of the original finish. Refer to Refinish instructions, Fig. 601, and to REPAIR-GEN for list of applicable standard practices.



REFINISH

PASSIVATE (F-17.09) AND CHROME PLATE (F-15.34)
0.0003-0.0005 THICK ALL OVER

REPAIR

125 MACHINE FINISH EXCEPT AS NOTED

MATERIAL: 15-5PH CRES 180-200 KSI

ALL DIMENSIONS ARE IN INCHES

Crosspin Refinish
Figure 601

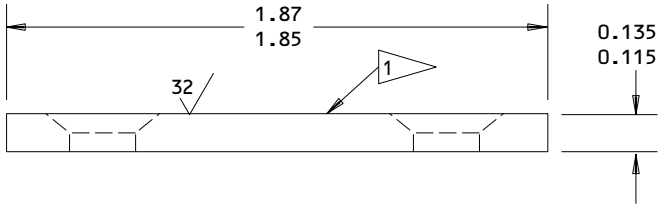
144225

LOCK SUPPORT PLATE - REPAIR 2-1

416T2812-10, -33

1. Coating Repair

   A. Repair is only replacement of the original finish. Refer to Refinish instructions, Fig. 601, and to REPAIR-GEN for list of applicable standard practices.



REFINISH

APPLY SOLID FILM LUBE ON AREAS NOTED 1. PASSIVATE OTHER SURFACES (F-17.09)

1 SOLID FILM LUBE BMS 3-8 (F-19.10)

REPAIR

125 ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

MATERIAL: 15-5PH CRES, 180-200 KSI (416T2812-10)
301 CRES, 1/2 HARD (416T2812-33)

ALL DIMENSIONS ARE IN INCHES

Lock Support Plate Refinish
Figure 601

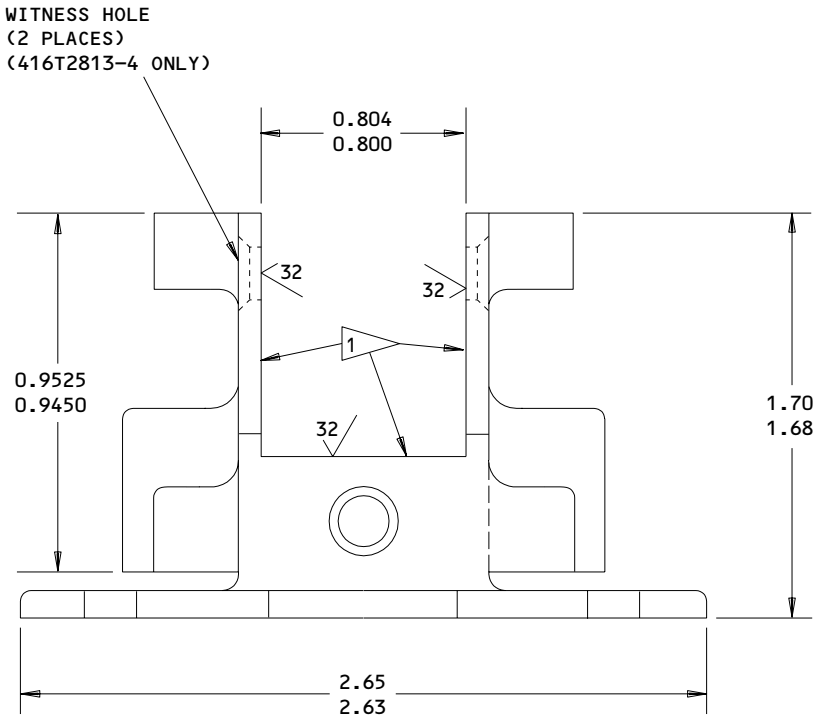
144224

BODY - REPAIR 3-1

416T2813-1, -3, -4

1. Plating Repair

NOTE: Repair consists of restoration of original finish. Refer to Refinish instructions, Fig. 601, and to REPAIR-GEN for list of applicable standard practices.



REFINISH

CHROME PLATE AND APPLY DRY FILM LUBE ON AREAS NOTED 1. PASSIVATE (F-17.09) OTHER SURFACES

1 CHROME PLATE (F-15.34), 0.0005-0.0010 THICK FOLLOWED BY DRY FILM LUBE BMS 3-8

REPAIR

125 MACHINE FINISH EXCEPT AS NOTED

MATERIAL: (416T2813-1) 17-4PH CRES, 180-200 KSI
(416T2813-3) 15-5PH CRES, 180-200 KSI

ALL DIMENSIONS ARE IN INCHES

Body Refinish
Figure 601

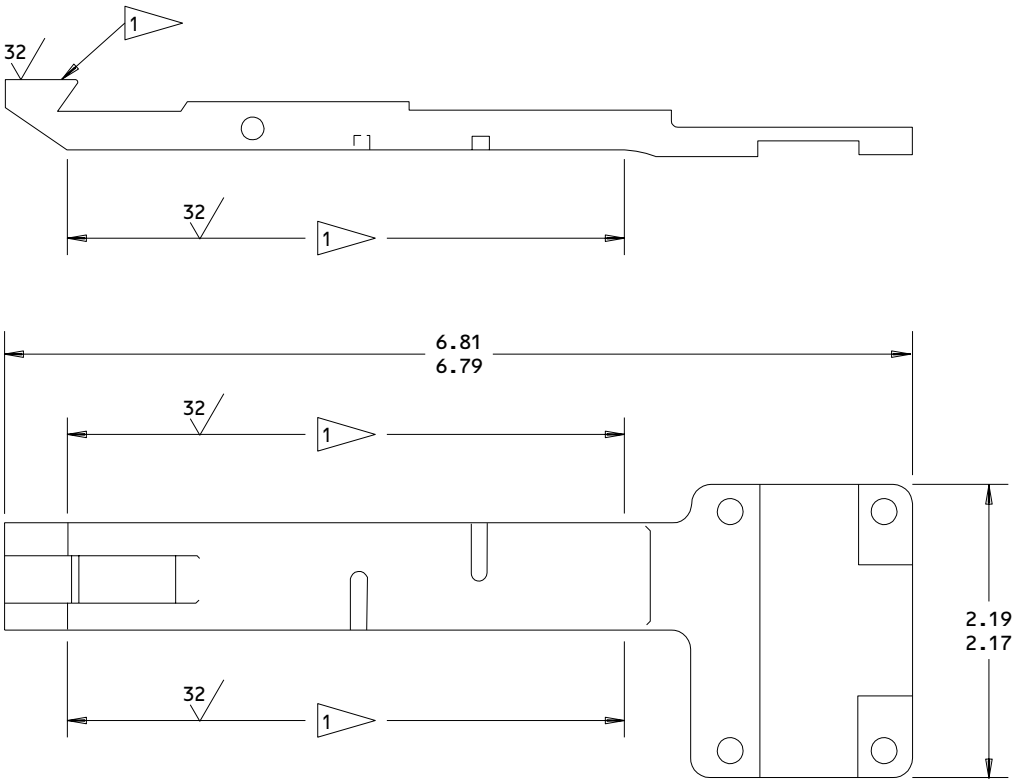
144226

CARRIER - REPAIR 4-1

416T2814-2, -6, -7, -9

1. Plating Repair

NOTE: Repair consists of restoration of original finish. Refer to Refinish instructions, Fig. 601, and to REPAIR-GEN for list of applicable standard practices.



REFINISH

CHROME PLATE AND APPLY DRY FILM LUBE ON AREAS NOTED 1. PASSIVATE (F-17.09) OTHER SURFACE.

1 CHROME PLATE (F-15.34), 0.0005-0.0010 THICK FOLLOWED BY DRY FILM LUBE BMS 3-8

REPAIR

125 MACHINE FINISH EXCEPT AS NOTED

MATERIAL: (416T2814-2,-7) 17-4PH CRES, 180-200 KSI
(416T2814-6,-9) 15-5PH CRES, 180-200 KSI

ALL DIMENSIONS ARE IN INCHES

Carrier Refinish
Figure 601

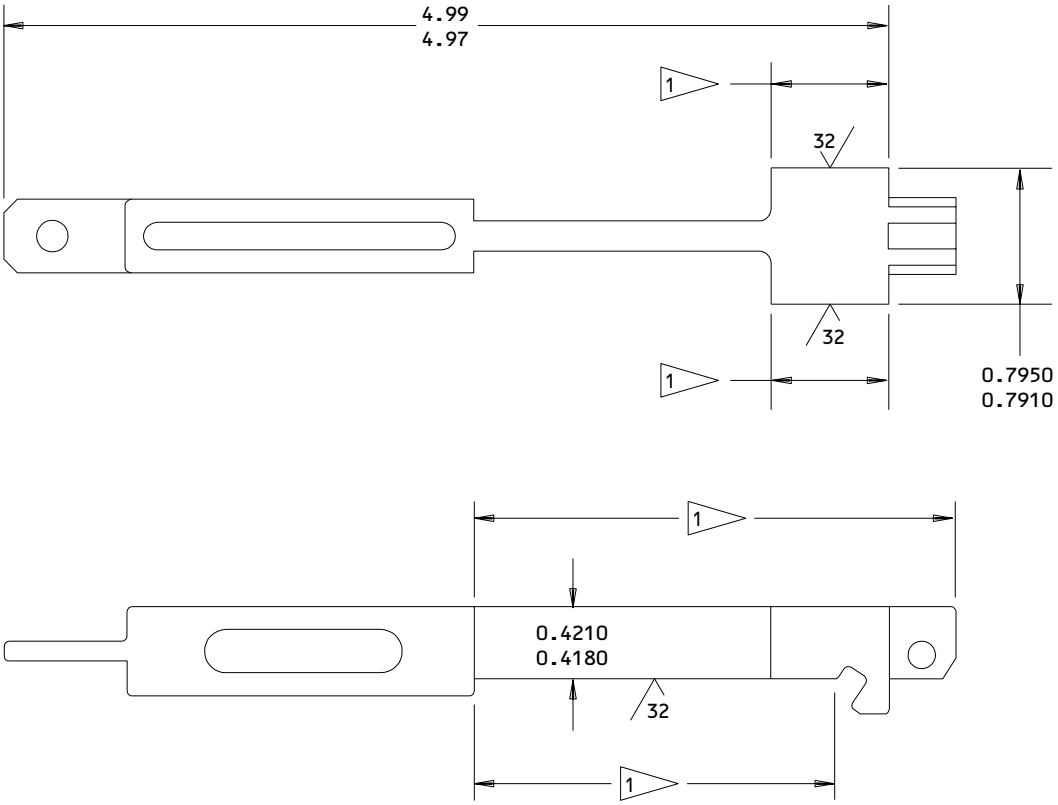
144227

CONNECTOR - REPAIR 5-1

416T2815-1, -3

1. Plating Repair

NOTE: Repair consists of restoration of original finish. Refer to Refinish instructions, Fig. 601, and to REPAIR-GEN for list of applicable standard practices.



REFINISH

CHROME PLATE AND APPLY DRY FILM LUBE ON AREAS NOTED 1. PASSIVATE (F-17.09) OTHER SURFACES

1 CHROME PLATE (F-15.34), 0.0005-0.0010 THICK FOLLOWED BY DRY FILM LUBE BMS 3-8

REPAIR

125 MACHINE FINISH EXCEPT AS NOTED

MATERIAL: (416T2815-1) 17-4PH CRES, 180-200 KSI
(416T2815-3) 15-5PH CRES, 180-200 KSI

ALL DIMENSIONS ARE IN INCHES

Connector Refinish
Figure 601

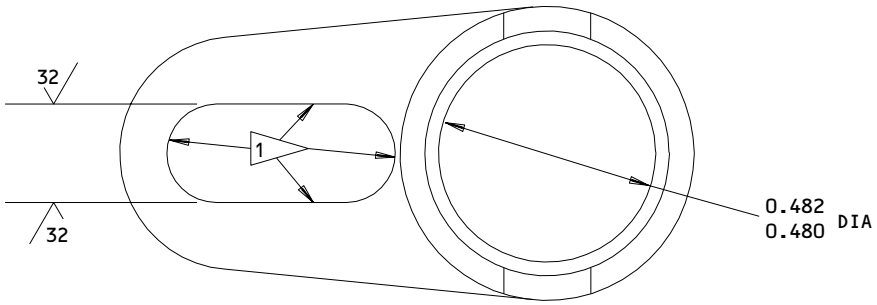
144228

CAM - REPAIR 6-1

416T2817-1, -3

1. Coating Repair

NOTE: Repair consists of restoration of original finish. Refer to Refinish instructions, Fig. 601, and to REPAIR-GEN for list of applicable standard practices.



REFINISH

APLLY DRY FILM LUBE ON AREAS NOTED 1.
PASSIVATE (F-17.09) OTHER SURFACES.

1 DRY FILM LUBE BMS 3-8

REPAIR

125 MACHINE SURFACE EXCEPT AS NOTED

MATERIAL: (416T2817-1) 17-4PH CRES, 180-200 KSI
(416T2817-3) 15-5PH CRES, 180-200 KSI

ALL DIMENSIONS ARE IN INCHES

Cam Refinish
Figure 601

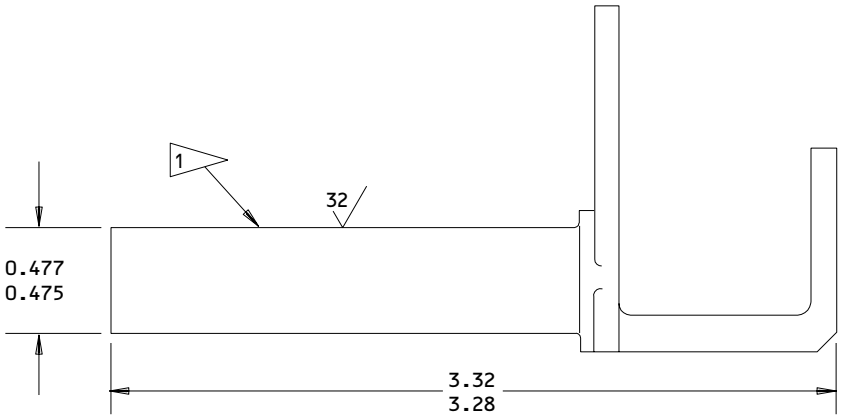
144229

BELLCRANK - REPAIR 7-1

416T2818-1, -3

1. Coating Repair

NOTE: Repair consists of restoration of original finish. Refer to Refinish instructions, Fig. 601, and to REPAIR-GEN for list of applicable standard practices.



REFINISH

APLLY DRY FILM LUBE ON AREAS NOTED 1.
PASSIVATE (F-17.09) OTHER SURFACES

1 DRY FILM LUBE BMS 3-8

REPAIR

125 MACHINE SURFACE EXCEPT AS NOTED

MATERIAL: (416T2818-1) 17-4PH CRES, 180-200 KSI
(416T2818-3) 15-5PH CRES, 180-200 KSI

ALL DIMENSIONS ARE IN INCHES

Bellcrank Refinish
Figure 601

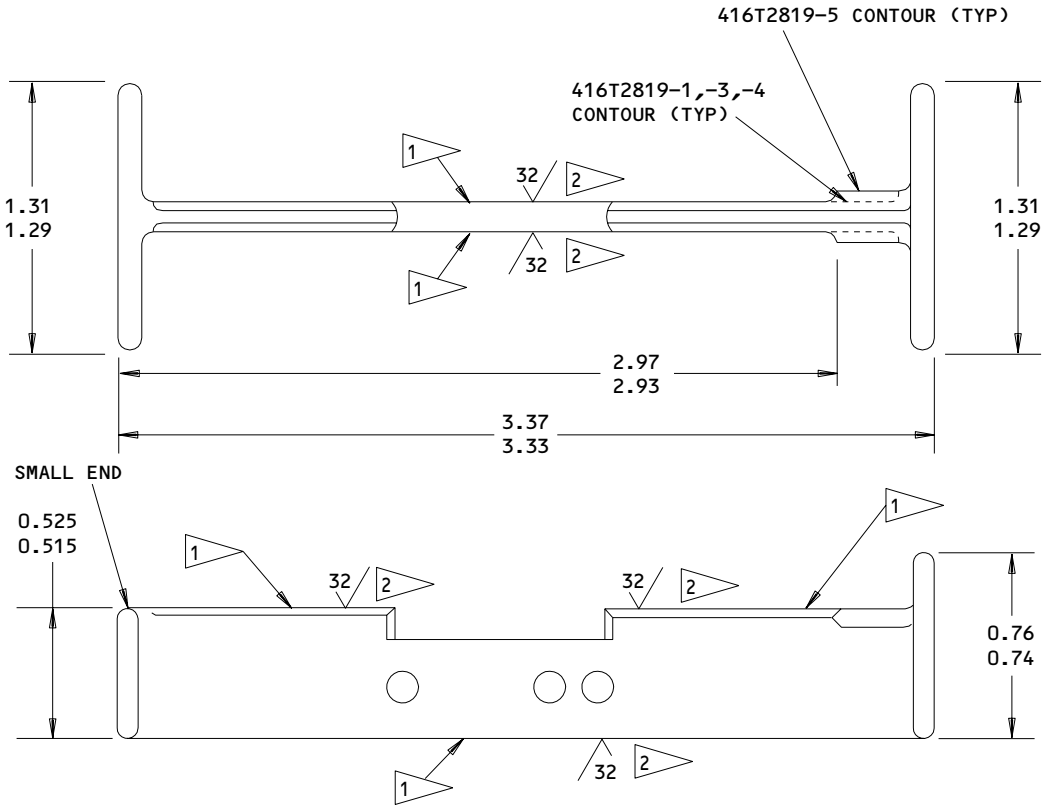
144230

LOCK - REPAIR 8-1

416T2819-1, -3, -4, -5

1. Coating Repair

NOTE: Repair consists of restoration of original finish. Refer to Refinish instructions, Fig. 601, and to REPAIR-GEN for list of applicable standard practices.



REFINISH

APPLY DRY FILM LUBE TO AREAS NOTED 1

1 DRY FILM LUBE BMS 3-8

2 416T2819-3 ONLY

REPAIR

416T2819-1,-4,-5: 32 ALL MACHINED SURFACES

416T2819-3: 125 ALL MACHINED SURFACES EXCEPT AS NOTED

MATERIAL:
(416T2819-1,-5) 17-4PH CRES, 180-200 KSI
(416T2819-3,-4) 15-5PH CRES, 180-200 KSI

ALL DIMENSIONS ARE IN INCHES

Lock Refinish
Figure 601

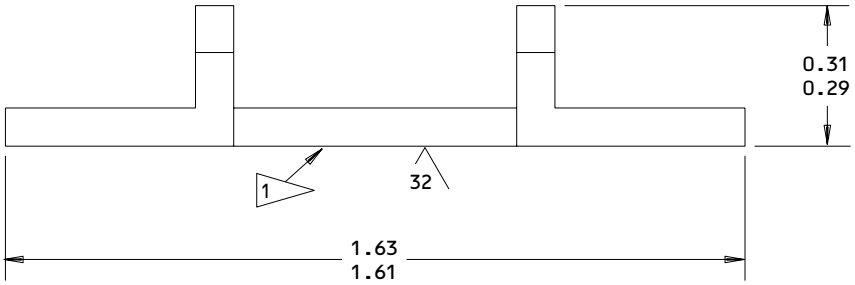
144231

CAP PLATE - REPAIR 9-1

416T2823-1

Coating Repair

NOTE: Repair consists of restoration of original finish. Refer to Refinish instructions, Fig. 601, and to REPAIR-GEN for list of applicable standard practices.



REFINISH

APPLY DRY FILM LUBE TO AREAS NOTED 1.
PASSIVATE (F-17.09) OTHER SURFACES

1 DRY FILM LUBE BMS 3-8

REPAIR

125 MACHINE SURFACE EXCEPT AS NOTED

MATERIAL: 17-4PH CRES, 180-200 KSI

ALL DIMENSIONS ARE IN INCHES

Cap Plate Refinish
Figure 601

144232

MISCELLANEOUS PARTS REFINISH - REPAIR 10-1

1. Repair of parts listed in Fig. 601 consists of restoration of original finish.

IPL FIG. & ITEM	MATERIAL	FINISH
Fig. 1		
Pawl (26), pins (27,28), screw (65)	15-5 PH CRES, 180-200 ksi	Passivate (F-17.09)
Hook (30) (416T2816-1)	15-5 PH CRES, 180-200 ksi	Passivate (F-17.09) all over
Hook (30A) (416T2816-2)	15-5 PH CRES, 180-200 ksi	Passivate (F-17.09) all surfaces except recessed areas. In recessed areas only: 1. Abrasive clean per 20-30-03. 2. Apply yellow primer BMS 10-11, type 1 (F-20.09). 3. Apply white enamel BMS 10-11, type 2 (F-21.17). 4. Apply two coats red-orange fluorescent paint Zynolyte 1419. 5. Apply clear protective finish type 41, except do not solvent clean.

Refinish Details
Figure 601

ASSEMBLY

1. Before installing pin (50) to fasten carrier in body, check hole in carrier for snug or slip fit with machined end lands of pin. This snug or slip fit is indication of satisfactory fit between carrier bore and knurled pin surfaces. Ream hole in carrier as necessary.

2. Temporarily install carrier in body and insert pin (50) through slot into body. Slide carrier back and forth to check for rubbing contact between pin and slot due to possible slight mismatch between slots and hole in mating parts. Check both sides of carrier and body. Remove material from slot edges as necessary. Pin should not rub on slot edges, or integrator will not function smoothly.

3. Install carrier in body. Press in pin (50) for equal protrusion on both sides. Recheck for rubbing and correct as necessary.

4. Before installing pin (27) in pawl (26), check hole in pawl for snug or slip fit with machined end lands of pin. This snug or slip fit is indication of satisfactory fit between pawl bore and knurled pin surfaces. Ream hole in pawl as necessary.

5. Install pawl (26) in connector (29) with pin (27).

6. Install hook (30) over pawl and lay assembled parts in carrier. Temporarily install plate (15) with screws (10) and check for free sliding of connector and hook. Remove interfering material from connector (29) or hook (30) as necessary; then install plate (15) on body with screws (10).

7. Slide a cam (55) onto bellcrank (40) shaft. Insert bellcrank through hole in body, and fit elongated hole in cam lug over end of pin (50). Rotate bellcrank so attach lugs point away from body and holes align between cam and bellcrank, then install pin (35). Slip remaining cam (55) over other end of bellcrank, turn to engage pin (50) in elongated hole; then install remaining pin (35).

8. Install latch lock (90) in body (120) so the smaller end face of the lock is on the same side of the unit as the attach lug of bellcrank (40) and the flange projection of carrier (105). Secure with lockplate (85) and screws (80).

9. Install ball (75), spring (70) and screw (65) in body (120).

ILLUSTRATED PARTS LIST

1. This section lists and illustrates replaceable or repairable component parts. The Illustrated Parts Catalog contains a complete explanation of the Boeing part numbering system.

2. Indentures show parts relationships as follows:

    Assembly
      Detail Parts for Assembly
      Subassembly
      Attaching Parts for Subassembly
        Detail Parts for Subassembly

    Detail Installation Parts (Included only if installation parts may be returned to shop as part of assembly)

3. One use code letter (A, B, C, etc.) is assigned in the EFF CODE column for each variation of top assembly. All listed parts are used on all top assemblies except when limitations are shown by use code letter opposite individual part entries.

4. Letter suffixes (alpha-variants) are added to item numbers for optional parts, Service Bulletin modification parts, configuration differences (Except left- and right-hand parts), product improvement parts, and parts added between two sequential item numbers. The alpha-variant is not shown on illustrations when appearance and location of all variants of the part is the same.

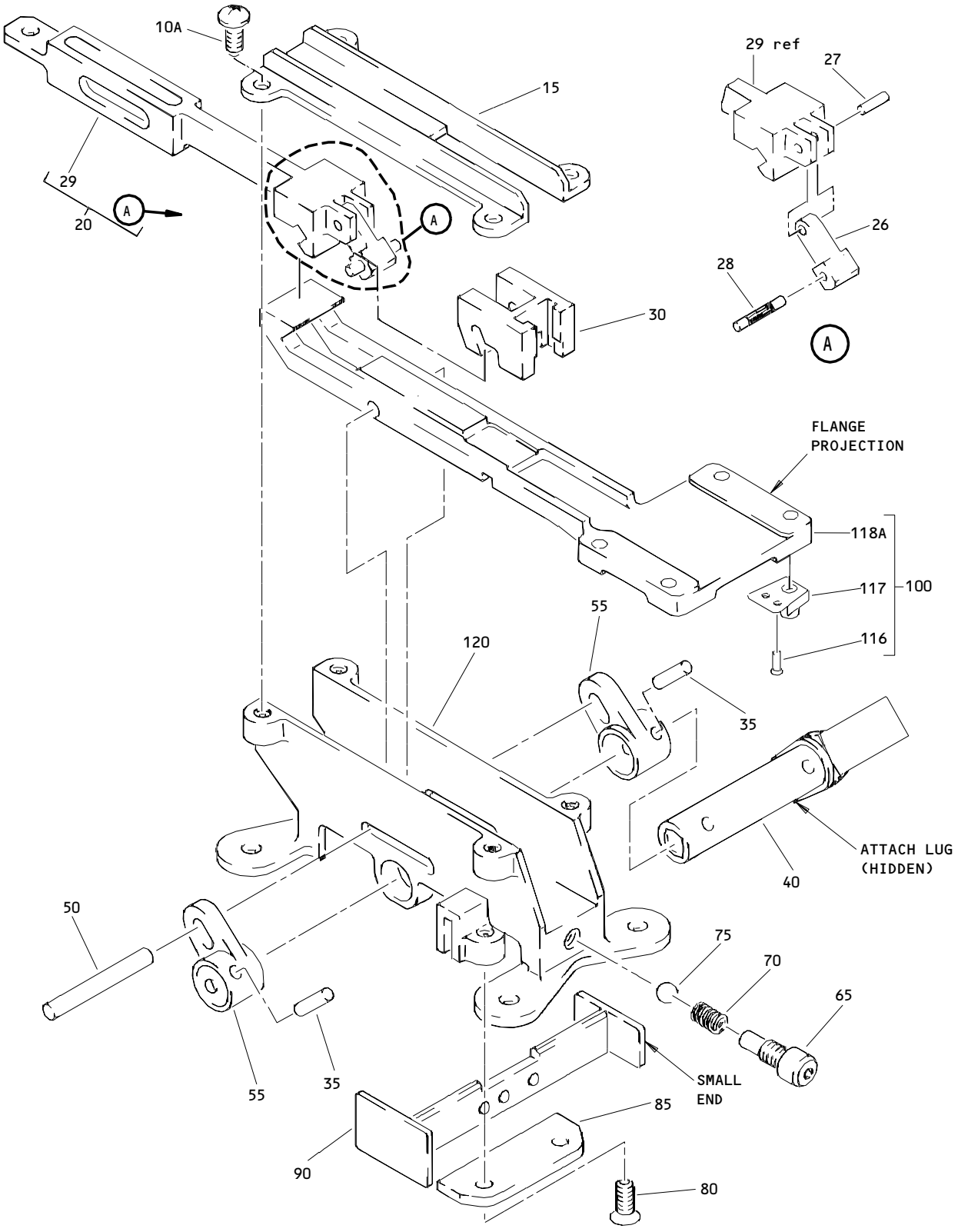
5. Service Bulletin modifications are shown by the notations PRE SB XXXX and POST SB XXXX.

    A. When a new top assembly part number is assigned by Service Bulletin, the notations appear at the top assembly level only. The configuration differences at detail part level are then shown by use code letter.

    B. When the top assembly part number is not changed by the Service Bulletin, the notations appear at the detail part level.

6. Parts Interchangeability

Optional (OPT)	The parts are optional to and interchangeable with other parts having the same item number.
Supersedes, Superseded By (SUPSDS, SUPSD BY)	The part supersedes and is not interchangeable with the original part.
Replaces, Replaced By (REPLS, REPLD BY)	The part replaces and is interchangeable with, or is an alternate to, the original part.



Offwing Escape System Integrator Assembly
Figure 1

U3990

FIG. & ITEM	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1234567	EFF CODE	QTY PER ASSY
01-					
-1	416T2811-1		DELETED		
-1A	416T2811-3		DELETED		
-1B	416T2811-5		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (LH) (POST SB 25-51) *[1]	A	RF
-1C	416T2811-7		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (LH) (PRE SB 25A0104)	C	RF
-1D	416T2811-9		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (LH) (PRE SB 25A0104)	E	RF
-1E	416T2811-11		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (LH)	G	RF
-1F	015T0449-5		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (LH) (POST SB 25A0104)	I	RF
-5	416T2811-2		DELETED		
-1G	015T0449-7		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (LH)	K	RF
-1H	416T2811-13		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (LH)	M	RF
-5A	416T2811-4		DELETED		
-5B	416T2811-6		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (RH)	B	RF
-5C	416T2811-8		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (RH) (PRE SB 25A0104)	D	RF
-5D	416T2811-10		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (RH) (PRE SB 25A0104)	F	RF
-5E	416T2811-12		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (RH)	H	RF
-5F	015T0449-4		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (LH) (POST SB 25A0104)	J	RF

FIG. & ITEM	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1234567	EFF CODE	QTY PER ASSY
01-					
-5G	015T0449-6		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (RH)	L	RF
-5H	416T2811-14		INTEGRATOR ASSY-OFF-WING ESCAPE SYS (RH)	N	RF
10	NAS623-3-1		DELETED		
10A	BACS12BG3P6		.SCREW (OPT ITEM 10B)		4
10B	BACS12BG3-6		.SCREW (OPT ITEM 10A)		4
15	416T2823-1		.PLATE-CAP		1
20	416T2812-1		.CONNECTOR ASSY	ABE-N	1
-20A	416T2812-2		.CONNECTOR ASSY (OPT)	EFK-N	1
-20B	416T2812-2		.CONNECTOR ASSY	CDIJ	1
26	416T2812-5		..PAWL		1
27	416T2812-7		..PIN-CONNECTOR		1
28	416T2812-8		..PIN-PAWL		1
29	416T2815-1		..CONNECTOR (USED ON ITEM 20)		1
-29A	416T2815-3		..CONNECTOR (USED ON ITEMS 20A,20B)		1
30	416T2816-1		.HOOK	A-F	1
-30A	416T2816-2		.HOOK	G-N	1
35	MS16562-236		.PIN-SPRING (OPT)		2
35A	MS51923-297		.PIN-SPRING (PREF) (OPT)		2
40	416T2818-1		.BELLCRANK	ABE-N	1
-40A	416T2818-3		.BELLCRANK (OPT)	EFK-N	1
-40B	416T2818-3		.BELLCRANK	CDIJ	1
50	416T2812-9		.PIN-CROSS		1
55	416T2817-1		.CAM	ABE-N	2
55A	416T2817-3		.CAM (OPT)	EFK-N	2
-55B	416T2817-3		.CAM	CDIJ	2
65	416T2812-11		.SCREW-DETENT		1
70	MS24585C40		.SPRING		1
75	BACB10TC2-12A		.BALL		1
80	NAS514P1032-5		DELETED		
80A	BACB30LU3P1		.BOLT	A-F I-L	2
-80B	BACS12BP3P6		.SCREW (OPT)	A-F I-L	2
-80C	BACB30LU3P2		.BOLT (OPT)	A-F I-L	2

FIG. & ITEM	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1234567	EFF CODE	QTY PER ASSY
01-					
80D	BACB30LR3P1		.BOLT	GHMN	2
80E	BACB30LH3P2		.BOLT	MN	2
			(OPT)		
85	416T2812-10		.PLATE-LOCK	A-F I-L	1
-85A	416T2812-10		.PLATE-LOCK *[2]	GHMN	1
-85B	416T2812-33		.PLATE-LOCK *[2]	GHMN	1
90	416T2819-1		.LOCK	AB	1
-90A	416T2819-5		.LOCK	E-H K-N	1
-90B	416T2819-3		.LOCK	CDIJ	1
			(OPT ITEM 95A)		
-90C	416T2819-4		.LOCK	CDIJ	1
			(OPT ITEM 95)		
100	416T2812-13		DELETED		
100A	416T2812-19		.CARRIER ASSY	A	1
100B	416T2812-19		.CARRIER ASSY	EGIKM	1
			(OPT ITEM 110B)		
105	416T2812-14		.CARRIER ASSY	BHJL	1
-105A	416T2812-14		.CARRIER	FLN	1
			(OPT ITEM 115A)		
-110	416T2812-15		DELETED		
110A	416T2812-21		.CARRIER ASSY	C	1
110B	416T2812-21		.CARRIER ASSY	EK	1
			(OPT ITEM 100B)		
115	416T2812-16		.CARRIER ASSY	D	1
-115A	416T2812-16		.CARRIER	FLN	1
			(OPT ITEM 105A)		
116	MS20427F3		..RIVET		8
			(SIZE DETERMINE ON INST)		
117	NS103225SE02		..NUTPLATE		4
			(V80539)		
			(SPEC BACN10KH3CM)		
			(OPT 109F9209M3		
			(V72962))		
			(OPT F29779-01-3		
			(V15653))		

FIG. & ITEM	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE 1234567	EFF CODE	QTY PER ASSY
01-					
118A	416T2814-2		..CARRIER (USED ON ITEMS 105, 105A)		1
-118C	416T2814-6		..CARRIER (USED ON ITEMS 115, 115A)		1
-118D	416T2814-7		..CARRIER (USED ON ITEMS 100A, 100B)		1
-118E	416T2814-9		..CARRIER (USED ON ITEMS 110A, 110B)		1
120	416T2813-1		.BODY	ABEF	1
-120A	416T2813-3		.BODY	C-FIJ	1
-120B	416T2813-3		.BODY (OPT)	CDIJ	1
-120C	416T2813-4		.BODY	GHK-N	1

*[1] REFER TO MANUAL 25-62-61 FOR PRE SB 25-51 CONFIGURATION
*[2] PLATE 416T2812-10 WITH SCREWS BACS12BP3P6 OR BOLTS BACB30LU3P2 OR BACB30LU3P1 OPTIONAL TO PLATE 416T2812-33 WITH BOLTS BACB30LR3P1